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FIGURE 1

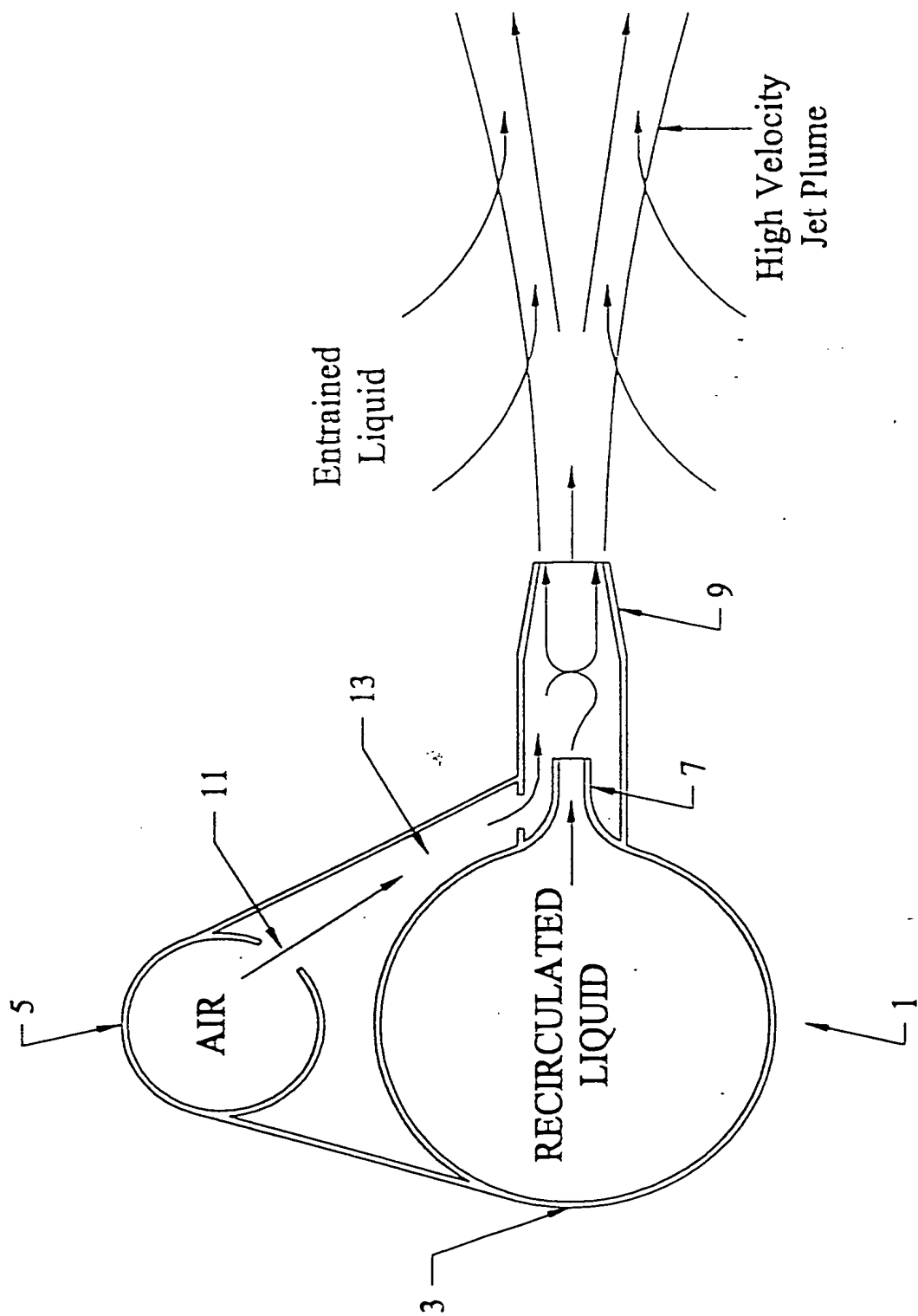
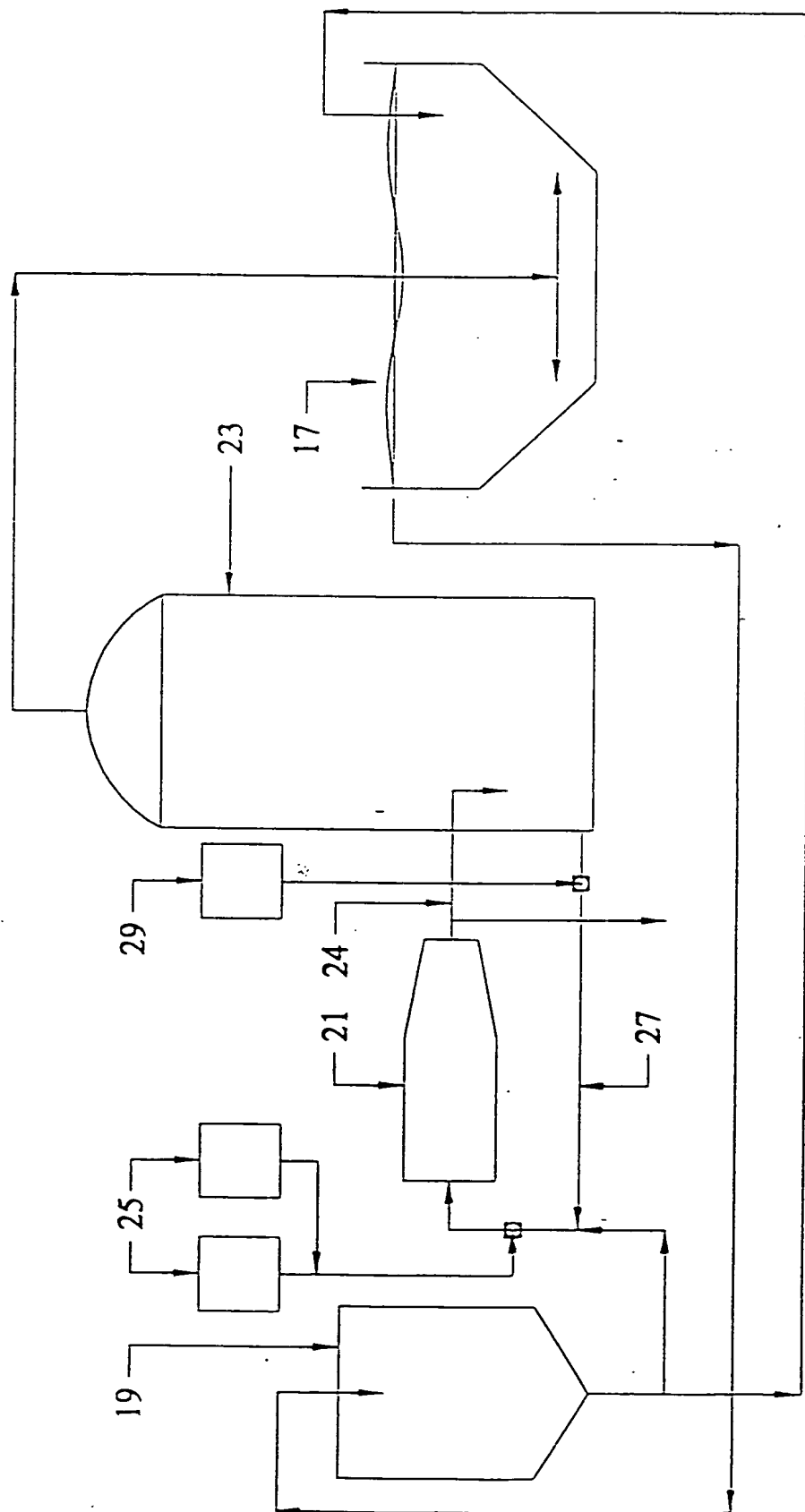
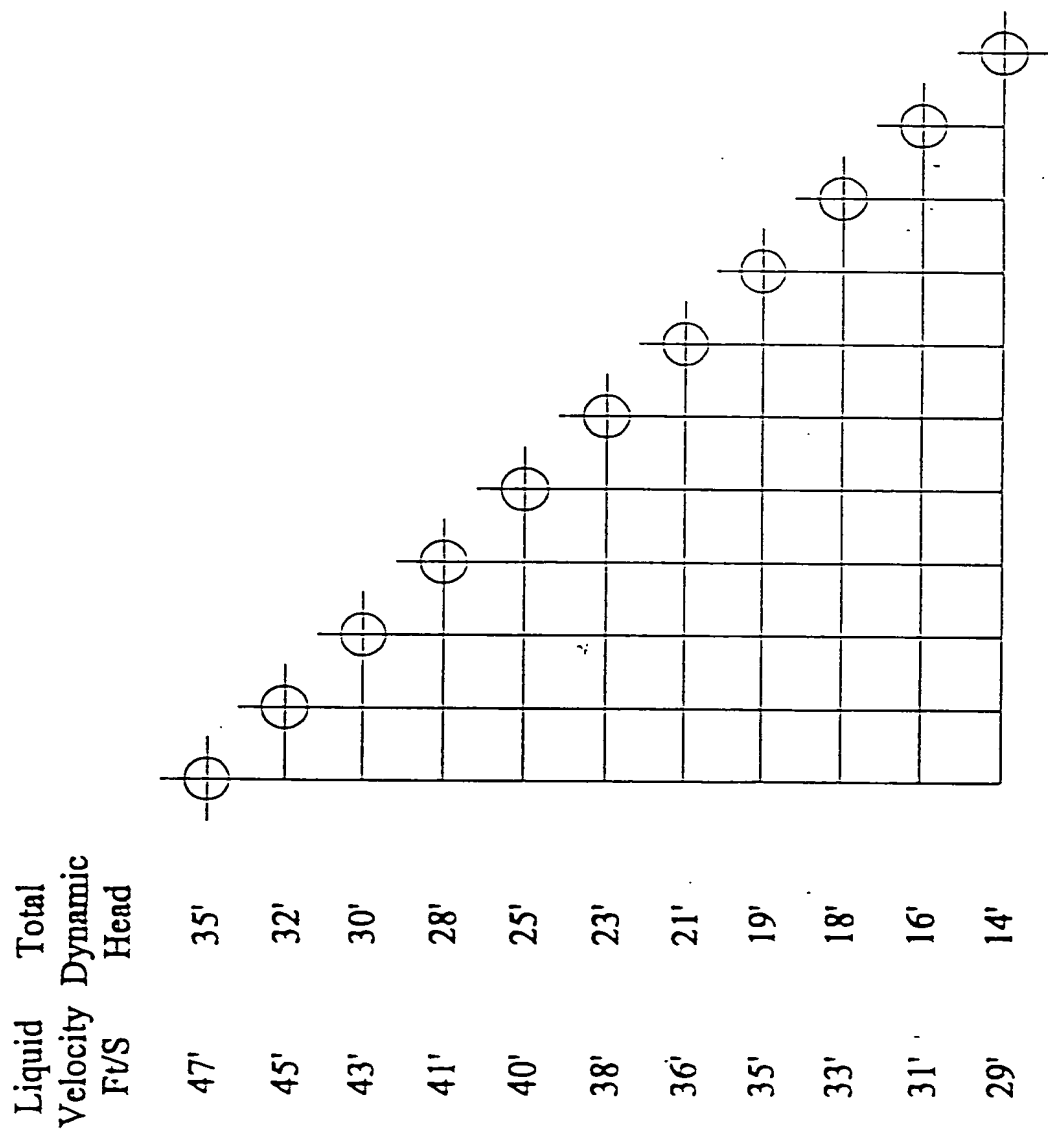


FIGURE 2



This schematic diagram illustrates a complex gas turbine engine system, likely for aircraft propulsion. The central component is a large cylindrical combustion chamber (33) containing three main sections: a compressor section (49), a combustor section (53), and a turbine section (55). A fuel supply system (27) feeds fuel into the combustor section (53) through a series of valves and pipes (28, 24, 57, 69). The turbine section (55) is mechanically coupled to a compressor section (49) via a shaft (41). The compressor section (49) draws air from the inlet (37) and compresses it before entering the combustor section (53). The exhaust gases exit the turbine section (55) through a nozzle (39). Various other components are shown, including a fuel control unit (35a, 35b), a fuel pump (37), and a fuel filter (39). The diagram also shows a fuel manifold (35) with multiple fuel lines leading to the combustor section (53).

FIGURE 4
Temperature Correlation Chart to RPM Velocity and TDH



Reactor Temperature °F 135 137 139 141 143 145 147 149 151 153 155
Pump RPM 770 740 710 680 650 620 590 560 530 500 470

FIGURE 5

